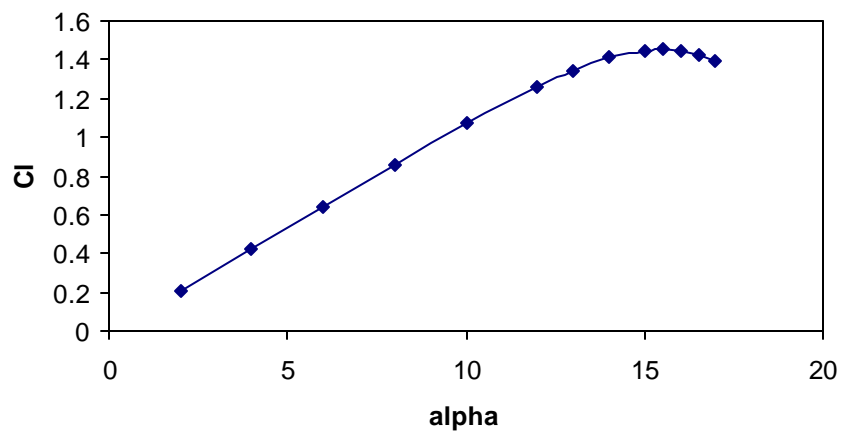
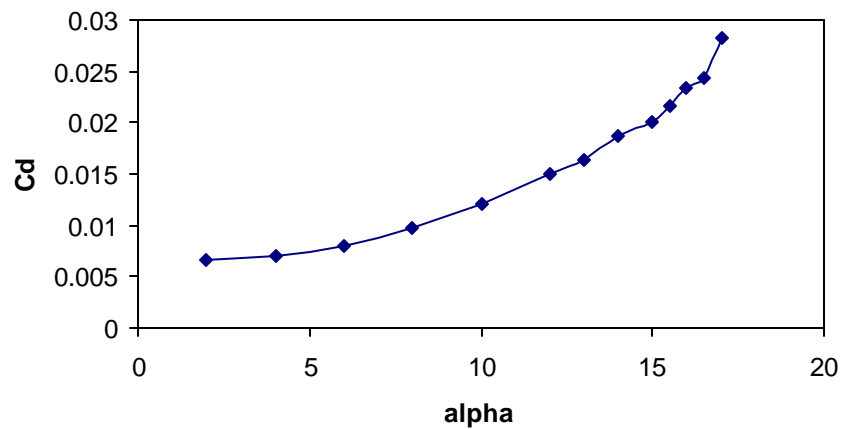


j	alpha	cl	cd	cm
1	2	0.21128	0.00665	-0.04984
2	4	0.42566	0.00695	-0.10098
3	6	0.64374	0.00793	-0.15344
4	8	0.86189	0.00977	-0.206
5	10	1.07121	0.01202	-0.25441
6	12	1.26185	0.01493	-0.29506
7	13	1.34311	0.0164	-0.30968
8	14	1.41254	0.01864	-0.32065
9	15	1.44717	0.02006	-0.31837
10	15.5	1.45074	0.0217	-0.31325
11	16	1.442	0.02332	-0.30547
12	16.5	1.42849	0.02439	-0.2996
13	17	1.39826	0.02818	-0.29315

Lift Coefficient, $RL=3E6$, $Mach=0.1$



Drag Coefficient, $RL=3E6$, $Mach=0.1$



Momentum Coefficient, $RL=3E6$, Mach=0.1

